NOTE: This message is not to be repeated or distributed.

FOR DISSEMINATION TO ALL SUBORDINATE UNITS:

ARMS FORCES/CGC/AGS DIVISION MD/AGM/AVN/L

INFO RUEFRGA/CGC/MHS/MOC/NGU-AGL/AL

SUBJECT: CLN TRP-100-33-25, APPENDIX C, CONVMENT THE UH-1H/LH

HELICOPTER HYDRAULIC SYSTEMS FROM MIL-H-5606 TO MIL-L-85292

HYDRAULIC FLUID (GEN-75-23)
A. REF A: VOIDE THE INSTRUCTIONS CONTAINED IN APPENDIX C OF

B. THE FOLLOWING INSTRUCTIONS ARE THE ONLY PROCEDURES TO BE

FOLLOWED WHEN CONVMENT THE UH-1H/LH HYDRAULIC SYSTEMS 95%

PAGE 2 ULNAC2721 UNCLAS

A. HYDRAULIC GROUND SERVICE UNIT, R/N NBU-1 OR D-9;

B. ITEMS TO BE MANUFACTURED:

A. DRAIN LINE FROM GROUND TEST COUPLING (RETURN) ON AIRCRAFT

C. REMOVAL OF COMPONENTS:

A. DISCONNECT ALL LINES TO THE HYDRAULIC PUMP AND ALLOW

LINES TO DRAIN; PLUS ALL LINES AFTER DRAINING, DISCONNECT AND

REMOVE THE HYDRAULIC PUMP FROM THE AIRCRAFT;

B. REMOVE THE FILTER: ELEMENT FROM THE HYDRAULIC MODULE AND ALLOW

MODULE TO DRAIN, CLEAN AND REINSTALL THE FILTER BOWL WITHOUT THE

FILTER ELEMENT;

C. COMPONENT FLUSHING:

A. TO FLUSH HYDRAULIC FLUID FROM THE HYDRAULIC PUMP, TURN THE

SPLINED SHAFT BY HAND WHILE PUMP IS INVERTED UNTIL FLUID NO LONGER

DRAINS FROM THE PRESSURE AND RETURN PORTS IN THE PUMP, FILL PUMP

THROUGH RETURN PORT WITH MIL-L-23282 AND REPEAT THE ABOVE PROCEDURE;

B. DISCONNECT ALL HYDRAULIC LINES TO THE MAIN ROTOR SERVOS

C. DISCONNECT THE CONTROL LINES FROM EACH CONTROL CYLINDER AT THE SWASHPLATE, ACTUATE THE

PISTON ON EACH CYLINDER FOUR FULL STROKES TO DISCHARGE RESIDUAL

FLUID FROM THE CYLINDER.
DISCONNECT ALL HYDRAULIC LINES TO THE TAIL ROTOR SERVO AT THE SERVO CYLINDER. DRAIN ALL LINES, DISCONNECT THE TAIL ROTOR SERVO CYLINDER AT THE CONTROL TUBE AND ACTIVATE THE PISTON FOUR FULL STROKES TO DISCHARGE RESIDUAL FLUID FROM THE CYLINDER.
D. DISCONNECT LINES TO GROUND TEST COUPLINGS (BOTH SIDES OF THE FIRE WALL) DRAIN LINES AND RECONNECT LINES TO THE COUPLINGS.
C.5. COMPONENT REPLACEMENT
A.1. INSTALL HYDRAULIC PUMP. FILL PUMP WITH MIL-H-83322 HYDRAULIC FLUID THROUGH THE CASE DRAIN PORT AND RECONNECT ALL HYDRAULIC LINES TO THE PUMP.
B.1. INSTALL PRESSURE AND RETURN LINES TO THE MAIN ROTOR SERVO CYLINDER AND RECONNECT CONTROL RODS TO THE SHIM PLATES.
CAUTION DO NOT INTERCONNECT HYDRAULIC HOSES FROM ONE CYLINDER TO THE OTHER.
C.1. INSTALL PRESSURE AND RETURN LINES TO THE MAIN ROTOR SERVO CYLINDER AND RECONNECT THE CONTROL ROD TO THE PISTON.
NOTE DO NOT REINSTALL FILTER ELEMENT AT THIS TIME.
C.6. FILLING, BLEEDING AND FUNCTIONAL TESTING.
REFER TO APPENDIX D OR E FOR SPECIFIC INSTRUCTIONS CONCERNING

PAGE 4 RULNAC2721 UNCLAS
CONVERSION OF GROUND SERVICING UNITS;
A. DISCONNECT THE RETURN HOSE TO THE RESERVOIR AT THE GROUND TEST COUPLING AND ALLOW TO DRAIN;
B. CONNECT THE DRAIN LINE TO GROUND TEST COUPLING (RETURN) AND PLACE UNATTACHED END INTO A TWO GALLON OPEN CONTAINER OR A CONTAINER MARKED TO INDICATE A TWO (2) GALLON LEVEL;
C. ATTACH PRESSURE HOSE FROM HYDRAULIC GROUND SERVICING UNIT TO THE GROUND TEST COUPLING (PRESSURE) ON THE AIRCRAFT;
D. BRING HYDRAULIC PRESSURE TO 1800 PSI AND METER HYDRAULIC FLUID INTO THE HYDRAULIC SYSTEM WHILE ACTUATING THE CYCLO Collective AND ANTI-TORQUE CONTROLS;
NOTE USE TWO MEN, IF NECESSARY, TO ACTUATE ALL CONTROLS SIMULTANEOUSLY.
E. OBSERVE FLOW OF HYDRAULIC FLUID FROM THE DRAIN LINE INTO THE TWO GALLON CONTAINER UNTIL CONTAINER IS FULL;
F. RELIEVE PRESSURE AT GROUND TEST UNIT, REMOVE PRESSURE LINE FROM AIRCRAFT AND SECURE HYDRAULIC SERVICE UNIT CAP GROUND TEST COUPLING (PRESSURE) ON AIRCRAFT;
G. REMOVE DRAIN LINE FROM GROUND TEST COUPLING (RETURN) AND CONNECT THE RETURN LINE FROM THE RESERVOIR TO THE GROUND TEST

PAGE 5 RULNAC2721 UNCLAS
COUPLING;
H. INSTALL A CLEAN FILTER ELEMENT, TORQUE AND SAFETY WIRE FILTER BOWL.
I. FILL RESERVOIR TO OVERFLOW USING MIL-H-8322 HYDRAULIC FLUID1
J. BLEED AND OPERATIONALY CHECK THE HYDRAULIC SYSTEM 7/A/X
TM55-1620-21D-20, CHAPTER 6, PAGE 6-8 PARA. I, J AND K.

1. FLY AIRCRAFT FOR ONE HOUR, DRAIN RESERVOIR AND ALL HYDRAULIC
   LINES TO THE HYDRAULIC PUMP, HYDRAULIC RESERVOIR, MAIN ROTOR SERVO
   CYLINDERS AND ANTI-TORQUE CYLINDER. REPEAT STEPS I AND J.

2. ACCOMPLISH FINAL FILLING AND BLEEDING DURING ENGINE RUN AT
   FLIGHT IDLE.

3. THIS NEW PROCEDURE WILL BE PUBLISHED AS A CHANGE TO
   TB55-1500-334-25 ASAP.

#2721.